Errata

Elastic buckling design curves for isotropic rectangular plates with continuity or elastic edge restraint against rotation

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The following corrections have been made to this paper published originally in the April issue, volume 104, number 1034 of *The Aeronautical Journal* on pages 175 to 182.

The Greek letter lambda (λ) was missing from the caption of Fig. 1; which should read:



Figure 1. A component plate: (a) of width *b* with loading and reference axis system and; (b) showing skew nodal lines with half-wavelength λ caused by perturbation force (denoted by *p* and *m*) and displacement amplitudes shown at the longitudinal edges of the plate, which are multiplied by exp(^{inx}/_λ).

Two multiplication symbols contained within the following sentence should have been printed as follows:

4.0 MODELLING

The isotropic material properties are: Young's Modulus (E) = 72.4kNmm⁻² (100,000lbin⁻²) and Poisson's ratio (v) = 0.3.

An S8R5 plate element is adopted for the finite element analysis⁽²⁰⁾ to provide the classical thin plate result. A high degree of convergence is achieved using 30×30 elements for the square plate ($\frac{4}{b} = 1.0$) and maintained by adjusting the number of elements with respect to changes in aspect-ratio, e.g. 45×30 elements for $\frac{4}{b} = 1.5$, etc.

The two equations contained in the paper should have appeared as follows:

$$\kappa = \frac{\beta b}{D} \qquad \dots (1)$$

$$\mathbf{k} = \frac{\sigma b^2 t}{\pi^2 \mathbf{D}} \quad \text{or} \quad k = \frac{\tau b^2 t}{\pi^2 \mathbf{D}} \qquad \dots (2)$$

The correct fig. 4(c) now appears below containing the "Critical shear load factor" as intended by the author, illustrating design curves that should correspond to Case 2-1:



4(c) longitudinal edges, y = 0 and b, the two transverse edges simply supported. Case 4-1 results⁽²⁹⁾, for longitudinal edges clamped, and Case 3-1 results⁽²³⁾, which possess continuity over simple supports in the *y*-direction, are given for comparison. Note that Case 4-1 $\Rightarrow k_{(a/b = w)} = 9.2$;